



HESSI SPACECRAFT DE-MATE CLOSEOUT

HSI_MIT_063A

2001-JUL-06

DAVE CURTIS

DRAFT

As Run on: _____ (Date/Time)

By _____ (Test Conductor)

DOCUMENT REVISION RECORD

Rev.	Date	Description of Change
A	2001-Jul-6	Original draft

Project Manager: _____
 Peter Harvey Date

System Engineer: _____
 David Curtis Date

QA: _____
 Ron Jackson Date

1. INTRODUCTION

1.1 Purpose

This document describes the physical configuration changes of the spacecraft following de-encapsulation of the launch vehicle fairing to safe the spacecraft and protect contamination-sensitive items. This list is divided into two parts: Those closeouts to be performed immediately following de-encapsulation, and those to be performed following de-mating the spacecraft from the launch vehicle.

2. POST DE-ENCAPSULATION CLOSEOUTS

This step is to be done immediately following de-encapsulation.

2.1 Items to Install

- 1. Fine Sun Sensor Cover Verify_____
- 2. CSS Dust Covers (8) Verify_____
- 3. RAS Dust Cover Verify_____
- 4. PMT Dust Cover Verify_____
- 5. SAS lens covers (3) Verify_____

QA Verify:

3. POST DE_MATE CLOSEOUTS

These must be done after the spacecraft is removed from the launch vehicle, and prior to powering up the spacecraft.

3.1 Items to Install

- 1. Spectrometer GSE via A/B Switch at Utility box Verify_____
- 2. Spectrometer LN2 feed GSE Verify_____

QA Verify:

3.2 Items to Remove

- 1. Actuator Enable Plug (de-mate) Verify_____
- 2. Spectrometer LN2 feed launch support Verify_____

QA Verify: